

Amendments To the Claims:

Please amend the claims as shown.

1. (currently amended) An ~~A~~annular combustion chamber (4) for a gas turbine (1), ~~wherein~~ whereby the annular combustion chamber (4) extends in an axial direction (A), encloses a combustor (7), and has on its inside facing the combustor (7) a bearing structure (26) on which a lining element (10) secured to this lines the annular combustion chamber (4), ~~characterized in that~~ wherein on the ~~a~~ rear side (13) facing away from the combustor (7) of two edge areas (15) ~~running~~ extending in the axial direction (A) on the lining element (10) a plurality of interlocking ~~means~~ devices (11) are located which have a ~~hook~~ width (B), and in that the lining element (10) is secured to the ~~corresponding~~ bearing structure (26) such that in order to release the lining element (10) from the bearing structure (26) the lining element (10) is moved by the extent of the ~~hook~~ width (B) of the interlocking means (11) in the axial direction (A).
2. (currently amended) An ~~A~~annular combustion chamber (4) according to Claim 1 ~~characterized in that~~, wherein a further plurality of interlocking ~~means~~ devices (11) are located as a central support (14) midway between two edge areas (15) of the lining element (10) running in the axial direction (A).
3. (currently amended) An ~~A~~annular combustion chamber (4) according to Claim 1, ~~or 2~~ ~~characterized in that~~ wherein two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (A) have a spacing (L) which is identical to or greater than the ~~hook~~ width (B) of the interlocking ~~means~~ devices (11).
4. (currently amended) An ~~A~~annular combustion chamber (4) according to one of the Claims 1, ~~to 3~~ ~~characterized in that~~ wherein each interlocking ~~means~~ device (11) has the identical hook width (B).

5. (currently amended) An Annular combustion chamber (4) according to ~~one of the Claims 1, to 4~~ characterized in that wherein two interlocking means (11) of the lining element (10) that are immediately adjacent in the axial direction (4) have a spacing (L) which is twice the hook width (B) of an interlocking means device (11).
6. (currently amended) An Annular combustion chamber (4) according to ~~one of the Claims 1, to 4~~ characterized in that wherein two interlocking means devices (11) of the lining element (10) that are immediately adjacent in the axial direction (4) have a spacing (L) which is three times the hook width (B) of an interlocking means device (11).
7. (currently amended) An Annular combustion chamber (4) according to Claim 5, ~~or 6~~ characterized in that wherein each spacing (L) between two interlocking means devices (11) of the lining element (10) that are immediately adjacent in the axial direction (A) is identical.
8. (currently amended) An Annular combustion chamber (4) according to ~~one of the Claims 1, to 7~~ characterized in that wherein the lining element (10) has stiffening ribs (12) running in the circumferential direction (U) of the annular combustion chamber (4) on its rear side (13) facing away from the combustor (7).
9. (currently amended) An Annular combustion chamber (4) according to Claim 8, characterized in that wherein the stiffening rib (12) is distanced from the interlocking means (11).
10. (currently amended) An Annular combustion chamber (4) according to ~~one of the Claims 1 to 7~~ characterized in that wherein the interlocking means (11) are L- and/or T-shaped.
11. (currently amended) A Gas turbine 1 with an annular combustion chamber (4) according to ~~one of the Claims 1 to 10~~ wherein the combustion chamber comprises:
a combustor enclosed within the chamber;
a bearing structure arranged on an inside facing of the chamber;

a lining element secured to the bearing structure and lining the chamber; and
a plurality of interlocking devices located on a rear side of two edge areas extending in
an axial direction on the lining element and having a width,
wherein the lining element is secured to the bearing structure such that in order to release the
lining element from the bearing structure the lining element is moved the width of the
interlocking device in the axial direction.

12. (new) A combustion chamber for a gas turbine, comprising:

a combustor enclosed within the chamber;
a bearing structure arranged on an inside facing of the chamber;
a lining element secured to the bearing structure and lining the chamber; and
a plurality of interlocking devices located on a rear side of an edge area extending in
an axial direction on the lining element and having a width,
wherein the lining element is secured to the bearing structure such that in order to release the
lining element from the bearing structure the lining element is moved the width of the
interlocking device in the axial direction.

13. (new) An annular combustion chamber according to Claim 12, wherein the interlocking
devices face away from the combustor.

14. (new) An annular combustion chamber according to Claim 12, wherein the interlocking
device is a hook.

15. (new) An annular combustion chamber according to Claim 12, wherein the width is the hook
width.

16. (new) An annular combustion chamber according to Claim 1, wherein the interlocking device
is a hook.

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17. (new) An annular combustion chamber according to Claim 1, wherein the width is the hook width.
18. (new) An annular combustion chamber according to Claim 2, wherein two interlocking means of the lining element that are immediately adjacent in the axial direction have a spacing which is identical to or greater than the hook width of the interlocking means.
19. (new) An annular combustion chamber according to one of the Claims 2, wherein each interlocking means has the identical hook width.
20. (new) An annular combustion chamber according to Claim 2, wherein two interlocking means of the lining element that are immediately adjacent in the axial direction have a spacing which is twice the hook width of an interlocking device.